In the Claims:

Please cancel claim 2 without prejudice.

Please amend claims 1, 3, 4, and 6-10 as follows:

1. (amended) A family of airfoils for a blade of a cooling-tower fan, wherein the blade has a root region and a tip region, the family of airfoils comprising an airfoil in the root region of the blade having a Reynolds number of 500,000, and an airfoil in the tip region of the blade having a Reynolds number of 1,000,000, wherein the airfoil in the tip region has a maximum lift coefficient of about 1.5 and the airfoil in the root region has a maximum lift coefficient of about 1.5, and wherein each airfoil is characterized by a maximum lift coefficient that is largely insensitive to roughness effects.

3. (amended) The family of airfoils of claim 1, wherein the blade has a length in a range of 3 to 10 meters.

4. (amended) The family of airfoils of claim 1, wherein the tip-region airfoil has a thickness of about 10% chord and the root region airfoil has a thickness of about 14% chord.

6. (amended) The airfoil of claim 4, wherein the blade is 3 to 10 meters in length.

7. (amended) An airfoil for a blade of a cooling-tower fan wherein the blade has a root region airfoil comprising an upper surface and a lower surface and a blade chord line wherein x/c values are dimensionless locations along the blade chord line and y/c values are dimensionless heights from the chord line to points on the upper or lower surface, wherein said values correspond substantially to the following table for said surfaces:

UPPER SURFACE

x/c	y/c
1.00000	0.00000
0.99662	0.00114
0.98703	0.00476

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0.97233	0.01078
0.95346	0.01852
0.93085	0.02701
0.90436	0.03546
0.87375	0.04370
0.83919	0.05188
0.80116	0.05998
0.76012	0.06785
0.71657	0.07535
0.67101	0.08232
0.62395	0.08859
0.57590	0.09397
0.52735	0.09831
0.47876	0.10147
0.43059	0.10333
0.38330	0.10381
0.33728	0.10284
0.29293	0.10039
0.25059	0.09648
0.21061	0.09119
0.17330	0.08462
0.13897	0.07691
0.10792	0.06822
0.08040	0.05875
0.05665	0.04869
0.03685	0.03828
0.02116	0.02780
0.00968	0.01758
0.00256	0.00808
0.00019	0.00179

LOWER SURFACE

x/c	y/c
0.00000	-0.00004
0.00021	-0.00165
0.00093	-0.00316
0.00215	-0.00470
0.00374	-0.00627
0.01354	-0.01266
0.02846	-0.01889
0.04821	-0.02465
0.07252	-0.02979
0.10113	-0.03414

0.13371	-0.03759
0.16991	-0.04003
0.20931	-0.04131
0.25153	-0.04120
0.29632	-0.03951
0.34353	-0.03619
0.39294	-0.03140
0.44418	-0.02524
0.49710	-0.01784
0.55160	-0.00978
0.60714	-0.00186
0.66285	0.00525
0.71775	0.01102
0.77079	0.01508
0.82084	0.01719
0.86679	0.01718
0.90735	0.01506
0.94113	0.01136
0.96729	0.00713
0.98565	0.00340
0.99645	0.00088
1.00000	0.00000.

8. (amended)   An airfoil for a blade of a cooling-tower fan wherein the blade has a tip region airfoil having a cross-sectional shape characterized by a thickness of about 10% chord and a maximum lift coefficient of about 1.5 to be substantially insensitive to roughness, and a Reynolds number of 1,000,000.

9. (amended)   The tip region airfoil of claim 8 wherein the blade is 3 to 10 meters in length.

10. (amended)   An airfoil for a blade of a cooling-tower fan wherein the blade has a tip region airfoil comprising an upper surface and a lower surface and a blade chord line wherein x/c values are dimensionless locations along the blade chord line and y/c values are dimensionless heights from the chord line to points on the upper or lower surface, wherein said values correspond substantially to the following table for said surfaces:

UPPER SURFACE

x/c	y/c
1.00000	0.00000
0.99670	0.00088
0.98716	0.00373
0.97222	0.00863
0.95269	0.01521
0.92905	0.02278
0.90137	0.03076
0.86962	0.03901
0.83410	0.04761
0.79539	0.05651
0.75405	0.06552
0.71067	0.07440
0.66582	0.08287
0.62009	0.09058
0.57397	0.09708
0.52766	0.10192
0.48128	0.10496
0.43504	0.10625
0.38928	0.10586
0.34435	0.10391
0.30064	0.10051
0.25854	0.09581
0.21849	0.08997
0.18089	0.08313
0.14614	0.07541
0.11457	0.06695
0.08648	0.05789
0.06211	0.04839
0.04163	0.03863
0.02516	0.02886
0.01280	0.01937
0.00455	0.01054
0.00047	0.00297
0.00003	0.00066

LOWER SURFACE

x/c	y/c
0.00004	-0.00070
0.00037	-0.00179
0.00120	-0.00266

0.00254	-0.00346
0.00771	-0.00536
0.02065	-0.00762
0.03926	-0.00898
0.06332	-0.00945
0.09261	-0.00909
0.12682	-0.00800
0.16562	-0.00627
0.20860	-0.00402
0.25530	-0.00138
0.30519	0.00152
0.35772	0.00455
0.41227	0.00755
0.46821	0.01041
0.52486	0.01296
0.58152	0.01510
0.63745	0.01667
0.69190	0.01759
0.74412	0.01779
0.79336	0.01725
0.83888	0.01593
0.87997	0.01390
0.91590	0.01120
0.94594	0.00809
0.96955	0.00501
0.98647	0.00240
0.99662	0.00063
1.00000	0.00000.